US Department of Transportation
Federal Aviation Administration

MAJOR REPAIR AND ALTERATION
(Airframe, Powerplant, Propeller, or Appliance)

Form Approved
OMB No. 2120-0020
For FAA Use Only
Office Identification

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed $1,000 for each violation (Section 901 of Federal Aviation Act of 1958).

1. Aircraft	Make: Ryan	Model: Navion B
	Serial No.: NAV-4-2313B	Nationality and Registration Mark: N5413K
2. Owner	Name *(As shown on registration certificate)*: Putney, William W III Rodgers, Gail C	Address *(As shown on registration certificate)*: 5780 Balmoral Drive Oakland, CA 94619

3. For FAA Use Only

The data identified herein complies with the applicable airworthiness requirements and is approved for the above described aircraft, subject to conformity inspection by a person authorized in FAR 43, Section 43.7

05/30/2003 (DATE) (SIGNATURE) OAK-FSDO

4. Unit Identification **5. Type**

Unit	Make	Model	Serial No.	Repair	Alteration
AIRFRAME	~~~~~~~~~~ *(As described in Item 1 above)* ~~~~~~~~~~				X
POWERPLANT					
PROPELLER					
APPLIANCE	Type				
	Manufacturer				

6. Conformity Statement

A. Agency's Name and Address	B. Kind of Agency	C. Certificate No.
Pierre Borduas 875A Island Dr. #253 Alameda, CA. 94502	X U.S. Certificated Mechanic Foreign Certificated Mechanic Certified Repair Station Manufacturer	A.P. 2020552 I.A.

D. I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

Date	Signature of Authorized Individual
6-2-03	P Borduas

7. Approval for Return To Service

Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is x APPROVED o REJECTED

BY							
		FAA Flt. Standards Inspector		Manufacturer	X	Inspection Authorization	Other *(Specify)*
		FAA Designee		Repair Station		Person Approved by Transport Canada Airworthiness Group	

Date of Approval or Rejection	Certificate or Designation No.	Signature of Authorized Individual
6-2-03	A.P. 2020552 I.A.	P Borduas

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

Description of alteration: This alteration is to replace a previously approved (337 dated 1/28/1946) upgraded non-structural instrument panel. The panel was upgraded to clean up and years of accumulated changes. This alteration provides a "Standard T" instrument layout which complies with 23.1321 "arrangement and visibility" and allows installation of upgraded instruments and avionics. (See attached drawing Fig 1)

Description of work: The new panel was fabricated using 6061-T6 0.100" in thickness. The new panel has been finished with a low reflectivity power coat finish. All required placards have been silk screened in a contrasting color in clear view of the pilot in compliance with 23.1541. A placard has been placed on the panel in compliance with 23.1301 "Function and installation" (b) adjacent to any equipment installed in the panel who's function is not clearly marked.

The new panel mounts using the pre-existing hardware, shock mounts (Lord PN: LD J6984-8) and mounting points used by the previous panel. The new panel does not significantly change the weight and balance of the aircraft.

A new weight and balance measurement in accordance with 43.13 chapter 10 has been done which includes this alteration.

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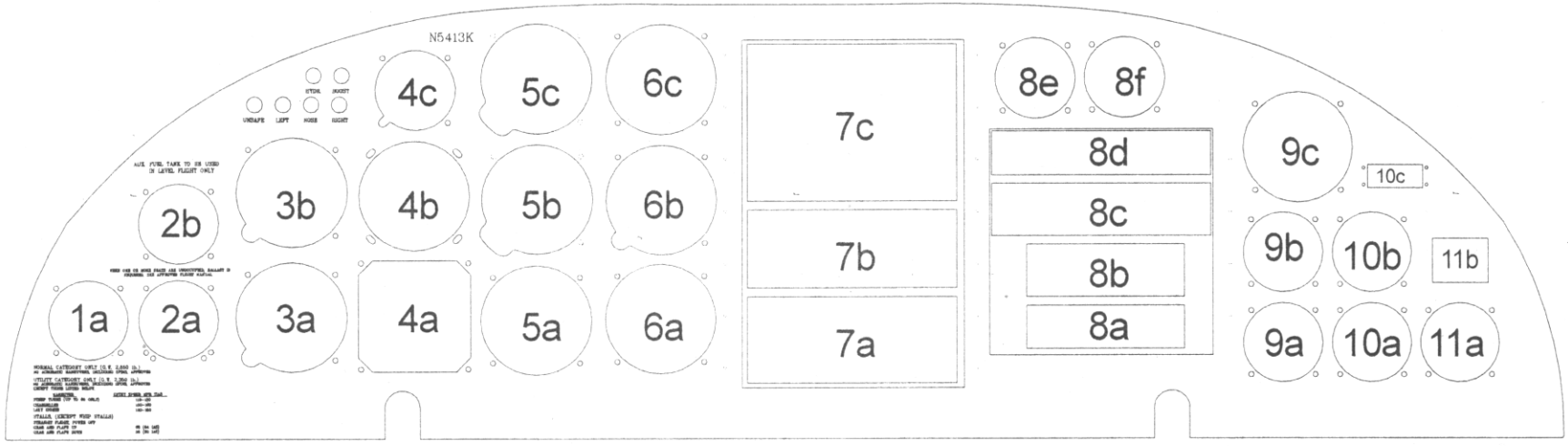
INSTRUCTIONS FOR CONTINUED AIRWORTHINESS

1) **Introduction:** See above (Form 337 section 8).
2) **Description:** See above (Form 337 section 8).
3) **Control:** Not applicable.
4) **Servicing information:** Not applicable.
5) **Maintenance Instructions:** Not applicable.
6) **Trouble shooting information:** Not applicable.
7) **Removal and replacement information:** The panel can be removed from the aircraft by disconnecting or removing the equipment installed in it and removing the 10-#10-32 flathead screws which secure it to the supporting structure.
8) **Diagrams:** Not applicable.
9) **Special inspection requirements:** Not applicable.
10) **Application of protective treatments:** Not applicable.
11) **Data:** No structural fasteners were used in the installation of this unit.
12) **List of special tools:** No special tools are required to install or maintain any components associated with this alteration.
13) **For commuter category aircraft:** Not applicable.
14) **Recommended overhaul periods:** Not applicable.
15) **Airworthiness Limitation Section:** Not applicable.

Revision: A letter will be submitted to the local FSDO with a copy of the revised FAA Form 337 and revised ICA. The FAA inspector accepts the change by signing Block 3 of the 337.

x Additional Sheets are Attached



Material: 6061-T6
Thickness: 0.100"
Finish: Low reflectivity powder coating

Reg: N5413K
SN: NAV-4-2313B

N5413K Panel

SIZE	FSCM NO.	DWG NO.	REV
A		NAV-2452781-13888	1.0

SCALE	DATE	SHEET
1:5	22 May, 2003	1 of 1