



US Department of Transportation

Federal Aviation Administration

## MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved

OMB No. 2120-0020

**For FAA Use Only**

Office Identification

**INSTRUCTIONS:** Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each violation (Section 901 of Federal Aviation Act of 1958).

<b>1. Aircraft</b>	Make <p style="text-align: center;">Ryan</p>	Model <p style="text-align: center;">Navion B</p>
	Serial No. <p style="text-align: center;">NAV-4-2313B</p>	Nationality and Registration Mark <p style="text-align: center;">N5413K</p>
<b>2. Owner</b>	Name (As shown on registration certificate) <p style="text-align: center;">Putney, William W III Rodgers, Gail C</p>	Address (As shown on registration certificate) <p style="text-align: center;">5780 Balmoral Drive Oakland, CA 94619</p>

**3. For FAA Use Only**

**4. Unit Identification**

**5. Type**

Unit	Make	Model	Serial No.	Repair	Alteration
AIRFRAME	----- (As described in Item 1 above) -----				<b>X</b>
POWERPLANT					
PROPELLER					
APPLIANCE	Type				
	Manufacturer				

**6. Conformity Statement**

<b>A. Agency's Name and Address</b>	<b>B. Kind of Agency</b>	<b>C. Certificate No.</b>
Pierre Borduas 875A Island Dr. #253 Alameda, CA. 94502	<input checked="" type="checkbox"/> U.S. Certificated Mechanic	A.P. 2020552 I.A.
	<input type="checkbox"/> Foreign Certificated Mechanic	
	<input type="checkbox"/> Certified Repair Station	
	<input type="checkbox"/> Manufacturer	

D. I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

Date <p style="text-align: center;">6-<del>8</del><sup>2</sup>-03<sub>MB</sub></p>	Signature of Authorized Individual <p style="text-align: center;"><i>P. Borduas</i></p>
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**7. Approval for Return To Service**

Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is  APPROVED  REJECTED

<b>BY</b>	FAA Flt. Standards Inspector	Manufacturer	<input checked="" type="checkbox"/>	Inspection Authorization	Other (Specify)
	FAA Designee	Repair Station		Person Approved by Transport Canada Airworthiness Group	

Date of Approval or Rejection <p style="text-align: center;">6-<del>8</del><sup>2</sup>-03<sub>MB</sub></p>	Certificate or Designation No. <p style="text-align: center;">A.P. 2020552 I.A.</p>	Signature of Authorized Individual <p style="text-align: center;"><i>P. Borduas</i></p>
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## NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

### 8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

**Description of alteration:** Installation of the J.P. Instruments model FS-450 Fuel Scan unit was accomplished in accordance with the J.P. Instruments "Fuel Flow Installation Manual".

**Description of work:** Installation of the indicator located at position 2b (see attached N5413K Panel) and the associated flow sensors was accomplished in accordance with STC SA00423SE and STC SA00861SE (See attachment). This installation was accomplished without deviation from the STC instructions.

A dedicated 3A fuse (Buss PN: GMA-3) labeled "FS-450" provides power the FS-450 indicator unit. The fuse is located in a block on the left lower side of the control panel. The total aircraft system electrical load does not exceed 80% of the generating capacity after this alteration.

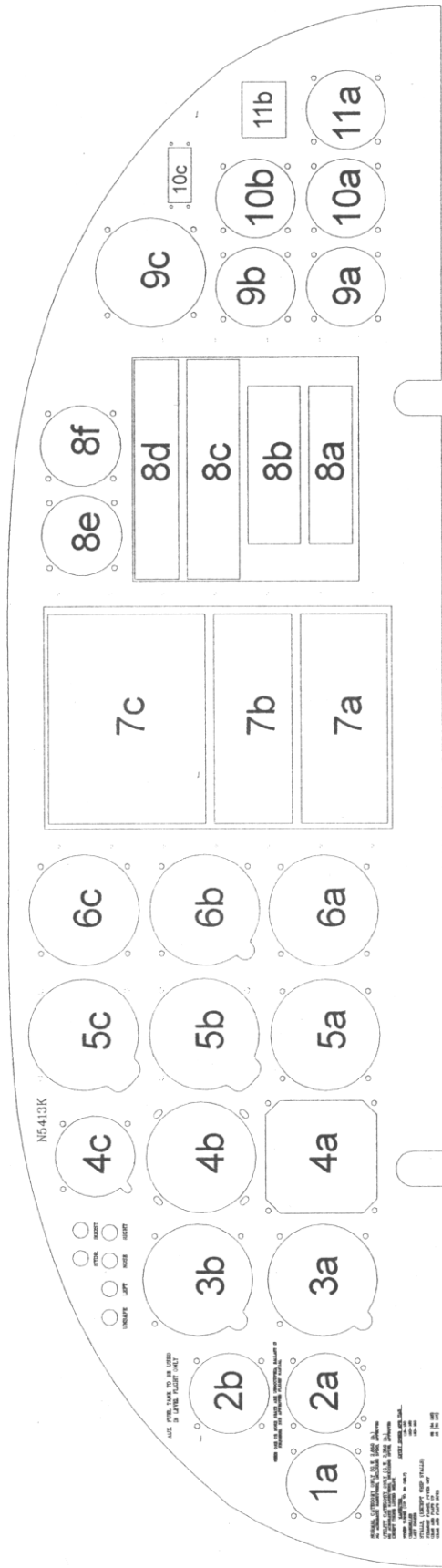
After installation a check for interactions with other systems was made to comply with 23.1309 "Equipment, systems and installations" (a)(1).

A new weight and balance measurement (conducted in accordance with AC 43.13 Chapter 10) has been done which includes this alteration.

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### INSTRUCTIONS FOR CONTINUED AIRWORTHINESS

- 1) **Introduction:** See above (Form 337 section 8).
- 2) **Description:** See above (Form 337 section 8).
- 3) **Control:** The manufacturer's flight manual supplement was inserted in the aircraft's POH. Additional operating instructions can be found in the J.P. Instruments "Fuel Scan FS-450 Pilot's Guide".
- 4) **Servicing information:** Not applicable.
- 5) **Maintenance Instructions:** Not applicable.
- 6) **Trouble shooting information:** Not applicable.
- 7) **Removal and replacement information:** Remove fuse to isolate the unit from aircraft power. Disconnect the D style connector from the rear of the unit. The instrument is attached to the panel with 4 #6-32 flat head screws. If the aircraft is to be returned to service without this unit installed, insure that cables and the connector are secured out of the way of flight controls.
- 8) **Diagrams:** Not applicable.
- 9) **Special inspection requirements:** Not applicable.
- 10) **Application of protective treatments:** Not applicable.
- 11) **Data:** No structural fasteners were used in the installation of this unit.
- 12) **List of special tools:** No special tools are required to install or maintain any components associated with this alteration.
- 13) **For commuter category aircraft:** Not applicable.
- 14) **Recommended overhaul periods:** Not applicable.
- 15) **Airworthiness Limitation Section:** Not applicable.
- 16) **Revision:** A letter will be submitted to the local FSDO with a copy of the revised FAA Form 337 and revised ICA. The FAA inspector accepts the change by signing Block 3 of the 337.



Material: 6061-T6  
 Thickness: 0.100"  
 Finish: Low reflectivity powder coating

Reg: N5413K  
 SN: NAV-4-2313B

# N5413K Panel

SIZE	FSCM NO.	DWG NO.	REV
A		NAV-2452781-13888	1.0
SCALE	1:5	DATE 22 May, 2003	SHEET 1 of 1